

IN THE SPECIFICATION:

Please correct the following paragraphs as they appear in the published version of this application, US Patent Application 2007/0090228 to read as follows:

– – **[0025]** The system 1 for protecting a civil aircraft 4 (see the drawing) from missiles 2 with infrared seeker heads 3 of portable anti-aircraft missile complexes comprises, on board the civil aircraft 4 being protected: launch sensors 5 ~~of the fact for sensing the launch of.~~ and launch coordinates 30 ~~for, the~~ ~~[[of]]~~ missile launch; a transceiver 6 having a turn drive 7 and an optical channel 8 which output 9 is connected to a sensor coordinate 10 ~~[[of]]~~ for continuously sensing the instantaneous missile coordinates ~~[[at]]~~ during the missile's flight on a missile flight trajectory 32; an on-board calculator 11; and a laser radiation generator 12 having an actuation device 13. The first group of inputs 14 of the on-board calculator 11 are connected to the outputs of sensors 5 ~~of the fact and coordinates of missile launch~~ in order to calculate the launch location coordinates 30 and the occurrence of the missile launch ~~place~~. The first output 15 of the on-board calculator 11 is connected to the turn drive 7 of the transceiver 6 in order to direct an input 16 of the optical channel 8 of the transceiver 6 to the launched missile 2. The output of the sensor 10 of instantaneous missile coordinates ~~[[at]]~~ of a missile flight trajectory 32 is connected to the second input 17 of the on-board calculator 11. The second output 18 of the on-board calculator 11 is connected to the actuation device 13 of the laser radiation generator 12. The aircraft 4 also has at least one engine 38 that radiates infrared power at a certain level.

[0026] It is desirable to provide the on-board calculator 11 with the third 19 and fourth 20 outputs coupled with the aircraft objective control system 34 ("black boxes" or flight recorders) and the aircraft system for communicating with the earth safety flight

providing services 36, respectively, in order to transmit the information on the fact of the missile launch and on the launch coordinates of missile launch ~~place~~ to the aircraft objective control system 34 and to the earth safety flight providing system 36. – –

– – **[0030]** A missile 2 with ~~[[the]]~~ infrared seeker head 3 could be launched ~~[[to]]~~ toward a civil aircraft 4 in the process of ~~[[its]]~~ flying. The sensors 5 ~~of the fact and~~ ~~coordinates of missile launch, disposed at~~ on the civil aircraft 4 being protected, ~~find out~~ sense the ultra-violet radiation of the engine of missile 2 being started. Signals from these sensors are fed to the first group of inputs 14 of the on-board calculator 11 for calculating the coordinates 30 of missile launch location ~~place~~. The information on the fact of this launch and coordinates of missile launch ~~place~~ are transmitted from the on-board calculator 11 via its third 19 and fourth 20 outputs to the aircraft objective control system 34 ("black boxes" or flight recorders) and to the aircraft system for communicating with the earth safety flight providing systems 36. Moreover, the control signal corresponding to the coordinates of missile launch is transmitted via the first output 15 of the on-board calculator 11 to the turn drive 7 of the transceiver 6 in order to direct the input 16 of the optical channel 8 of the transceiver 6 ~~[[to]]~~ toward the launched missile 2. Through this optical channel 8 the infrared radiation of the flying missile 2 comes to the sensor 10 of missile coordinates ~~[[at a]]~~ on the missile flight trajectory 32, which is ~~[[the]]~~ a narrow-directed sensor of the infrared range, and as a result, an output signal of this sensor is formed. After processing the output signal of the sensor 10 ~~of missile coordinates at a missile flight trajectory~~ in the on-board calculator 11 , the instantaneous coordinates of the missile 2 in ~~the~~ at any given time ~~moment~~ are calculated. – –